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Category II High Temperature Evaluation of a T-38 Aircraft

Clair K. Sandstrom Billy L. White

FLIGHT TEST DIVISION
FLIGHT AND ENGINEERING TEST GROUP

MAY 1961





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Clair K. Sandstrom
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Flight and Engineering Test Group

May 1961

Systems Project Number 420L

Aeronautical Systems Division
Air Force Systems Command
United States Air Force
Wright-Patterson Air Force Base, Ohio

FOREWORD

This report on the T-38A High Temperature Evaluation was prepared by Clair K. Sandstrom and Billy L. White of the Flight Test Engineering Branch, Flight and Engineering Test Group, Wright Air Development Division (WADD), at the request of the Directorate of Systems Management, Airborne Support Systems Division, T-38 Support Systems Project Office under Systems Project Number 420L. Mr. Sandstrom was project engineer; Capt Russell L. Rogers, Air Force Flight Test Center, was the project pilot.

The high temperature tests were conducted at the El Centro Naval Auxiliary Landing Field (NALF), California, from 19 August through 5 September 1960.

The Flight and Engineering Test Group, WADD, directed the tests, provided the engineering and instrumentation personnel, and was responsible for final data reduction and analysis, and publication of the test results. Data for preliminary instrument flight procedures for the Flight Manual were also gathered during the desert tests and will be published in a separate report by the Flight Test Engineering Branch. Operational information influenced by desert climates is presented in paragraph 25 of "Recommendations" as proposed additions to Section IX of the applicable flight handbook, T.O. T-38A-1.

The Air Force Flight Test Center (AFFTC), Edwards Air Force Base, California, provided the project pilot, supply support, and maintenance personnel. The El Centro NALF provided limited base support.

Technical representatives of Norair Division, Northrop Aircraft Corporation, Hawthorne, California, and General Electric Corporation participated in the test. The T-38A ''Talon'' (USAF S/N 59-1590) aircraft was designed and manufactured by Norair; instrumentation pick-ups and leads were installed by Northrop during production in accordance with military specification MIL-I-5289(ASG) as amended by WADD. The aircraft was powered by two YJ85-GE-5 eight-stage axial-flow turbojet engines. Additional instrumentation and the necessary recording equipment were installed in the aircraft at Edwards Air Force Base, California, prior to the El Centro tests, by WADD instrumentation personnel, with shop support furnished by AFFTC.

ABSTRACT

The T-38A aircraft was subjected to flight and static tests at ambient temperatures up to 115°F at the El Centro Naval Auxiliary Landing Field, California. No serious deficiencies were encountered. The cockpit temperatures were unsatisfactory; however, the refrigeration package in the test aircraft was not a production unit and is to be replaced by a more efficient unit in the production aircraft.

Tests proved maintenance of the aircraft in a desert climate presents no unusual problem, but strict adherence to operational procedures, as affected by high temperatures, is necessary.

PUBLICATION REVIEW

This report has been reviewed and approved for publication.

FOR THE COMMANDER:

IOSEPH DA**Y**(S, JR.

Colonel, USA

Commander, Plight and Engineering

Test Group

REVIEW AND APPROVAL OF WADD TECHNICAL NOTE 60-283

Author Approval

CLAIR K. SANDSTROM

T-38A Project Engineer

Extreme Operating Conditions Section

Flight Test Engineering Branch

Section and Branch Concurrences

Edwin C. Brown Jr. EDWIN C. BROWN, JR., Cappain, USAF

Chief, Extreme Operating Conditions Section

Flight Test Engineering Branch

RAY C. GORDON, JR., Major, USAF

Chief, Flight Test Engineering Branch

Flight Test Division

Division Concurrence

Webster Wflound WEBSTER W. PLOURD, Lt Colonel, USAF

Chief, Flight Test Division

Group Approval

HUGH'S, LIPPMAN

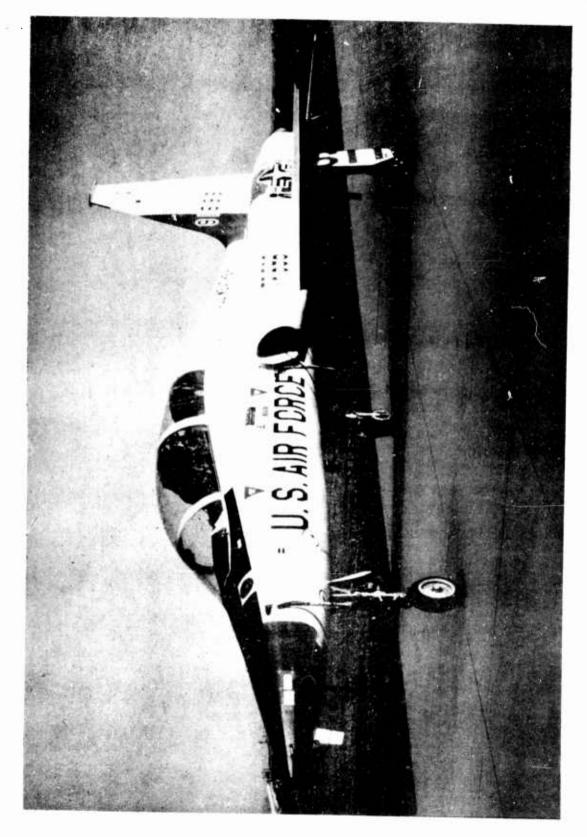
Technical Director

Flight and Engineering Test Group

WADD TN 60-283

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INTRODUCTION

GENERAL

- 1. High temperature tests of a T-38 "Talon" aircraft (USAF S/N 59-1596), Figure 1 (frontispiece), were conducted at El Centro Naval Auxiliary Landing Field, El Centro, California. Objectives of the tests were to:
 - a. Determine the adequacy of the aircraft systems and components for functioning under high temperature climatic conditions.
 - b. Analyze high temperature deficiencies discovered and make recommendations for corrective action.
 - c. Determine desert operating procedures to be included in Section IX of the applicable flight handbook, T. O. T-38A-1.
 - d. Compile environmental data for design purposes.
- 2. Flight tests were designed to subject the aircraft to the most extreme temperature conditions (ambient temperatures to 115°F) which may reasonably be expected while operating in desert climates. Ground tests were conducted to obtain data in two different areas, (1) to determine the effects of high temperatures and solar heat on the aircraft, its systems and components, and (2), to determine the maximum temperature during normal engine checkouts.
- 3. Systems and components of the aircraft presented no problems which can be directly or indirectly attributed to high temperatures.

DESCRIPTION OF THE TEST AIRCRAFT

- 4. The T-38A aircraft is a twin engine, two-place tandem trainer designed for speeds in the supersonic range. The mission of the aircraft is to accomplish all phases of basic pilot training, including day and night transition, formation, navigation, and adverse weather flying. The T-38A employs a low, thin wing; a movable, horizontal stabilizer; and fuselage lines characterized by distinct reverse curvature in the area of the wing root in conformance with the "area rule" concept. The crew normally consists of a student pilot and an instructor; the aircraft can be flown solo from either cockpit.
- 5. The test aircraft had a basic weight of approximately 7760 pounds with an internal fuel capacity of 3640 pounds. For this test, the weight of the fully loaded aircraft, two crew members, 3640 pounds of fuel, and instrumentation recording equipment, was 11,800 pounds.
- 6. The YJ85-GE-5 engines, mounted nearly parallel to each other and near the center line of the aircraft, comprise an axial-flow eight-stage compressor coupled directly to

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a two-stage turbine. The engine incorporates a variable area inlet-guide vane system, a controlled compressor interstage bleed, an afterburner with a variable area exit nozzle, and employs a direct impingement type starting system.

TEST INSTRUMENTATION

- 7. Basic instrumentation included most of the thermocouples and transducers for the temperature and pressure pick-ups with all the necessary wiring leading from the pick-ups to the rear cockpit. The recording equipment, installed in the rear cockpit, included six switch boxes, a balance box, a junction box, and a digitizer equipped with a tape transport for recording. Most of the equipment was placed in the rudder pedal wells, which permitted an observer in the aft cockpit for all test flights. (The control stick and rudder pedals had been removed prior to the delivery of the aircraft to facilitate the installation of the recording system).
- 8. The flight test engineer controlled the recording system from a control panel mounted in the rear cockpit. During engine starts, the controls were set on "continuous" and only the first two switch boxes were used to get engine start data since they contained all of the engine parameters. This gave a repeating cycle every four seconds for any one parameter. When all six of the 30 position switch boxes were used, each parameter was sampled once every 12 seconds since the digital recording system has the capability of recording 180 parameters in 12 seconds. Throughout the test, the recording system was cycled at one-minute intervals during stabilized conditions, and operated continuously during transient conditions.
- 9. A total of 119 parameters including 77 temperatures, 28 pressures, and 14 miscellaneous measurements were recorded by the digital recording system. A complete listing of the instrumentation is included in Appendix A.
- 10. Solar radiation was measured by an Epply pyrheliometer coupled to a modified Brown milli-volt recorder. The pyrheliometer is sensitive to those factors such as cloud coverage, moisture, and dust, which affect the amount of solar heat reaching the test aircraft. Mounted on the platform adjacent to the pyrheliometer were ten thermocouples which included one shaded, two with black metal discs, and seven unshielded. (See Figure 2 of Appendix C for the exact locations.) The thermocouples were positioned to record the ambient temperatures around the test aircraft from ramp level up to six feet above the ramp.

TEST PROCEDURES AND RESULTS

FLIGHT TEST SUMMARY

Foreign Object Damage

- 11. The engines are started with an external air supply. Engine starting is accomplished by direct impingement of low-pressure air on the second-stage turbine wheel of the two-stage engine rotor.
- 12. Periodic inspection subsequent to the desert test showed that the turbine sections of both engines had suffered foreign object damage. No damage whatsoever was discovered in the compressor sections. It appeared that either the small baffles from the

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first-stage turbine or foreign objects had entered the turbine section causing the damage. The foreign objects may have been forced into the first stage of the turbine through the starter air ducts. As the engine started motoring, air from the engine compressor would have forced the objects back through both sections of the turbines. The objects damaged turbine nozzles, blades, and partitions. On the day before the final ground engine run, a violet wind storm had occurred, in which quantities of sand and pebbles were blown around extensively. Some of these may have entered the duct of the MA-1 starting unit. As it is not common practice to inspect or blow out this duct before attaching the MA-1 unit to the starter air duct receptable, it is likely that the foreign objects entered the engines in this manner.

Accessory Power Shaft Failure

13. Immediately after the flight on 22 August, a turn-around flight had to be abandoned because of a failure in the hydraulic system due to shearing of the radial shaft in the engine driven accessory gearbox. (The engine mounted accessories are driven from an accessory gearbox on the under side of the engine. The gearbox is driven by a radial shaft from the engine rotor. One of the drive pads on the accessory gearbox of each engine drives an airframe mounted accessory power assembly which consists of a twospeed automatic shifting gearbox. Each gearbox drives a generator and hydraulic pump. The two-speed gearboxes shift automatically during engine acceleration or deceleration to maintain a safe operating speed for the generators and hydraulic pumps.) When the left engine was started the generator caution light remained on and there was no hydraulic pressure in the left hydraulic system. Subsequent investigation revealed that the radial shaft (P/N 741C-872) in the engine driven accessory gearbox had sheared. The malfunction apparently had been caused by the failure of the two-speed gearbox to shift down during engine shut-down after the previous flight. The failure was not a high temperature fault and has happened frequently on other T-38 aircraft. The gearbox was removed during the periodic inspection and thoroughly examined. There were no faults or maladjustments found. The contractor has had serious quality control problems with this particular vendor and has changed vendors.

GROUND TEST SUMMARY

Static Temperature Survey

- 14. The static temperature surveys were conducted during two weekend periods. The first, on 20-21 August, was compromised to some extent by light winds and clouds. The second period, on 26, 27, 28 August, represented more extreme conditions. Selected data from the latter test are presented in Appendix C. The solar radiation and ambient air temperatures are plotted for all three days as shown in Figures 3 through 6 of Appendix C.
- 15. For the static temperature survey, the aircraft was parked on the ramp away from buildings or other obstructions which would cast shadows. The aircraft was parked with its longitudinal axis oriented in the north-south direction; cockpit canopies were closed and locked for the entire heat-soak period to obtain the maximum cockpit temperatures under the most extreme conditions. Temperature data were recorded at 30-minute intervals from 0700 to 1800 hours and at 60-minute intervals from 1800 to 0700 hours. No covers were used on the canopies.
- 16. Analysis of the data presented in Appendix C shows that the maximum temperature of 160°F occurred in the upper fuselage and in the cockpit area where the effects of solar radiation are greatest.

- 17. Comparison of the data over the complete soak period indicates that little or no heat is retained in the aircraft or aircraft systems from one day to the next. Therefore, opening or closing the aircraft in the evening will have no effect on the peak temperatures on the following day.
- 18. No unusual problems are anticipated if the aircraft is kept closed during high temperature periods. It would be advisable, however, to vent the aircraft when weather conditions permit, to reduce the temperatures within the aircraft. Normal high temperature problems associated with increased deterioration of rubber materials, plastics, and lubricants may be expected along with minor thermal stress.

Ground Engine Run

19. The engines were operated at various stabilized speeds from idle to maximum power with afterburners to determine the maximum temperatures which may be expected to occur in the aircraft compartments and systems during normal desert ground engine checkouts. Although high temperatures were reached, none of them appeared to be excessive. The engine run data does not appear in this report but is available from the originator upon request.

CONCLUSIONS

20. No serious deficiencies were encountered during the tests that could be attributed to high temperature operations. Maintenance of the aircraft in a desert climate presented no unusual problems; therefore, the aircraft may be considered suitable for desert-type operation such as encountered in the test area.

RECOMMENDATIONS

- 21. The following recommendations are based on the results of the high temperature functional evaluation tests conducted on the T-38A aircraft at El Centro NALF:
- a. Eliminate the possibility of foreign objects entering the starting unit hose by placing a screening device in the adapter between the starting unit hose and the starter air duct receptacle of the engine. (Reference paragraph 12).
- b. Include the Desert Operating Procedures, which follow, in Section IX of the applicable flight handbook, T. O. T-38A-1.

DESERT AND HOT WEATHER PROCEDURES

22. Hot weather and desert procedures differ from normal procedures mainly in that additional precautions must be taken to protect the aircraft from damage caused by high temperatures and dust. Particular care should be taken to prevent the entrance of sand into the various aircraft components and systems (engine, fuel system, pitot-static systems, etc.). All filters should be checked more frequently than under normal conditions. Units with plastic or rubber parts should be protected as much as possible from wind-blown sand and excessive temperatures. Canopy covers should be left off to prevent sand from entering between the cover and the canopy and acting as an abrasive on the plastic.

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Heat buildup under closed canopy will not damage cockpit at runway temperatures of 110°F or less.

Before Entering Aircraft

NOTE: All metal surfaces exposed to the sun will be extremely hot to the touch, and gloves should be worn during preflight inspections.

- 1. Check exposed portions of shock strut pistons for dust and sand and have them cleaned if necessary.
- 2. Check inflation of shock struts and hydraulic accumulators which may have become over-inflated due to temperature increases.
 - 3. Check tires carefully for improper inflation and blistering or cord separation.
 - 4. Be sure all protective covers are removed from aircraft.
 - 5. Check intake duct for accumulations of dust or sand.
- 6. Make sure all filters have been cleaned and that the aircraft has been thoroughly inspected for fuel or hydraulic leaks caused by the swelling of packings or expanding of fittings.
- 7. Inspect area behind aircraft to make sure sand or dust will not be blown onto personnel or equipment during starting operations.
 - 8. Check starting unit duct for accumulations of small stones and sand.

On Entering Aircraft

- 1. Check cockpit for accumulation of dust or sand.
- 2. Check instruments and controls for moisture from high humidity, and apply heat if necessary, to dry them.

NOTE: All items in the cockpit that have been exposed to the sun are extremely hot to touch unless the cockpit has been protected by a canopy sun shade.

Ground Operation

- 1. Normal starting procedures are used in hot weather.
- 2. Expect the engines to accelerate more slowly than on a normal or cold day.
- 3. Ground checking should be complete but accomplished as rapidly as possible.
- 4. Limit use of brakes as much as possible during taxi operation to prevent overheating.

Takeoff

CAUTION: Takeoff planning is very important with high ambient temperatures and runways of marginal length. Takeoff distance will be longer because the air is less dense and ground speed will be increased for the same indicated airspeed. The refusal speed and distance, and the speed and distance checks should be carefully calculated.

- 1. To prevent additional drag caused by excessive angle of attack, the nose should not be lifted off until slightly below the recommended rotation speed.
- 2. Hot weather operation requires the pilot to be cautious of gusts and wind shifts near the ground.

During Flight

1. During high altitude flight the windshield and canopy defrost system should be operated at the highest possible temperature consistent with the crew members' comfort. This will preheat the transparent surfaces, thus precluding the formation of frost or fog during descent.

Approach and Landing

- 1. Check very closely and maintain recommended indicated approach and touchdown speeds. Because of high outside air temperatures, speed relative to the ground is higher than normal.
 - 2. Anticipate a long landing roll due to higher ground speed at touchdown.
 - 3. Use all available runway for stopping to avoid overheating the brakes.

Before Leaving Aircraft

1. Make sure that the protective cover is installed immediately on the pitot boom.

NOTE: Engine intake and exhaust duct covers should not be installed before normal engine cool-down period.

APPENDIX A

INSTRUMENTATION SUMMARY

The T-38A test aircraft's basic instrumentation was installed during production by the Norair Division of the Northrop Corporation in accordance with military specification MIL-I-5289(ASG) as amended by WADD. The aircraft was instrumented under the concept of selective instrumentation in order to keep the total number of pick-ups to a minimum.

Essentially, the instrumentation recorded data from the left engine, the hydraulic system, the air-conditioning system, the fuel system, the electrical system, and flight parameters. A total of 119 parameters including 77 temperatures, 28 pressures, and 14 miscellaneous measurements were recorded by the digital recording system mounted in the rear cockpit of the aircraft.

The data were recorded at various rates depending on the information required. During engine starts the controls were set on "continuous" and only the first two switch boxes were used to get engine "start" data, with a repeating cycle for any one parameter every four seconds. When all six of the 30-position switch boxes were used, each parameter was sampled once every 12 seconds. Throughout the test, during stabilized conditions the recording system was cycled at one-minute intervals, and during transient conditions the system was operated continuously.

A list of the instrumentation parameters, with switch box positions and corresponding ranges, is included in Table 1 of this appendix.

Table 1
SWITCH BOX INSTRUMENTATION

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 1 (Pressure)	
1		Counter-Cycle	
2		Counter-Time	
3		Potentiometer-Time	
4	1	Airspeed	
5	2	Altitude	
6	3	Outside Air Temp.	
7	4	Exhaust Gas Temp	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 1 (Pressure) (Cont'd)	
8	8	Tachometer Speed	
9	9	Throttle Position	0-115°
10	5	Main Fuel Flow Left-Hand Engine	(LB/HR)
11	11	Static Reference Manifold Pressure	± 6 PSID
12	400	Compressor Air Inlet Pressure (11:00 Position)	± 12. 5 PSID
13	401	Compressor Air Inlet Pressure (7:00 Position)	± 12.5 PSID
14	405	Fuel Pressure at Inlet to Left-Hand Fus. Fuel Filter	± 25 PSID
15	406	Fuel Pressure at Outlet to Left-Hand Fus. Fuel Filter and Inlet to Engine and Afterburner Fuel Pumps	± 35 PSID
16	407	Fuel Pressure at Main Eng. Fuel Manifold Line	0-1000 PSIG
17	402	Oil Press into Eng. Pump (Oil Tank Pressure)	± 6 PSID
18	4 03	Oil Pressure Out of Eng. Pump	0-75 P S IG
19	10	Cabin Conditioning Control Valve Position	0-90° 1000
20	420	Static Air Sensor (Differential Pressure Between Cockpit and Static Air Sensor)	± 6 PSID
21	422	Static Pressure of Bleed Air Downstream of Shutoff Valve	± 2.5 PSID
22	423	Total Pressure of Bleed Air Downstream of Shutoff Valve	0-150 PSIG
23	424	Static Pressure of Conditioned Air Downstream Hot-Cold Junction ${\rm P}_{12S}$	of ± 2.5 PSID

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 1 (Pressure) (Cont'd)	
24	425	Total Pressure of Conditioned Air Downstream of Hot-Cold Junction P ₁₂ T	0-30 PSIA
25	426	Static Pressure of Bleed Air By-Pass P _{10S}	± 2.5 PSID
26	427	Total Pressure of Bleed Air By-Pass $^{ m P}$ 10T	0-100 PSIG
27	428	Static Pressure; Bleed Air Defog Duct Upstream Nozzle (Fwd. C/P Windshield)	± 25 PSID
28	429	Static Pressure at Cockpit Ram Air in Valve	± 12.5 PSID
29		Bridge Power Reference	
		BOX NR 2 (Temperature)	
1	50	Compressor Air Inlet Temp. (11:00 Position)	
2	51	Compressor Air Inlet Temp. (7:00 Position)	
3	54	Fuel Temp. Center of Fwd. Fus. Tank	
4	55	Fuel Temp, at Outlet of Fwd. Fuel Cell Boost Pump	
5	56	Fuel Temp, at Outlet of Left-Hand Fus, Fil. and Inlet to Engine	
6	58	Fuel Temp, at Fuel Oil Cooler Outlet	
7	59	Fuel Temp, at Main Eng. Fuel Flowmeter and Inlet to Oil Cooler	
8	52	Oil Temp. into Engine	
9	53	Oil Temp. out of Engine (Nr 3 Bearing)	
10	77	Amb. Air Temp. Adjacent to Cabin Temp. Sensor	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	ANGE
		BOX NR 2 (Temperature) (Cont'd)	
11	78	Cockpit Air out Downstream of Equip. Ram Air Junction	
12	79	Defog Air Temp. Upstream of Defog Nozzle	
13	80	Bleed Air Temp. Downstream of Shutoff Valve	
14	81	Bleed Air at By-Pass T ₁₀ (Control Valve Inlet)	
15	82	Air Temp. Downstream of Moisture Sep. Anti- Ice Valve T ₁₁	
16	83	Air Temp. Downstream of Hot-Cold Junction T_{12}	
17	76	Cockpit Air in Temp. (Fwd.)	
18	85	Surface Temp. Base of Windshield (Center)	
19	85A	Surface Temp. Base of Windshield (Left Side)	
20	88	Surface Temp. Left-Hand Center of Windshield (Center)	
21	88A	Surface Temp. Left-Hand Center of Windshield (Left Side)	
22	98	Surface Temp. Center of Windshield (Top)	
23	98A	Surface Temp. Left Side of Windshield (Aft)	
24	86	Surface Temp. Top Center of Fwd. Canopy	
25	89	Surface Temp. Left Center of Fwd. Canopy	
26	89A	Surface Temp. Left Bottom of Fwd. Canopy	
27	87	Surface Temp. Top Center of Aft. Canopy	
28	90	Surface Temp. Left Center of Aft Canopy	
29	90A	Surface Temp. Left Bottom of Aft Canopy	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 3 (Temperature)	
1	70	Pilot's Foot Temp	
2	71	Pilot's Waist Temp. (Unshielded)	
3	71A	Pilot's Waist Temp. (Shielded)	
4	7 2	Pilot's Head Temp.	
5	73	Student's Foot Temp	
6	74	Student's Waist Temp. (Unshielded)	
7	74A	Student's Waist Temp. (Shielded)	
8	75	Student's Head Temp.	
9		Zero Reference	
10	91	Ambient Temp. Aft Electrical Compartment (STA-282) Right-Hand	
11	92	Ambient Temp. Aft Electrical Compartment (STA-282) Left-Hand	
12	93	Ambient Temp. Aft Electrical Compartment (STA 277)	
13	94	Ambient Temp. Fwd. Electrical Compartment (STA 103.5) Right-Hand	
14	95	Ambient Temp, Fwd, Electrical Compartment (STA 112) Right-Hand	
15	96	Ambient Temp, Fwd. Electrical Compartment (STA 135) Left-Hand	
16	97	Ambient Temp. Fwd. Electrical Compartment (STA 135) Right-Hand	
17	110	Utility Pump Inlet Oil Temp.	
18	111	Utility Pump Outlet Oil Temp.	
19	112	Utility Reservoir Inlet Oil Temp.	
20	113	Utility Cooler Inlet Oil Temp.	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 3 (Temperature) (Cont'd)	
21	114	Flt. Control Pump Inlet Oil Temp.	
22	115	Flt. Control Pump Outlet Oil Temp.	
23	116	Flt. Control Reservoir Inlet Oil Temp.	
24	117	Flt. Control Inlet Oil Temp., Right-Hand Rudder Cyl.	
25	118	Flt. Control Outlet Oil Temp., Right-Hand Rudder Cyl.	
26	119	Flt. Control Cooler Inlet Oil Temp.	
27	120	Skin Temp, to Left-Hand Aileron Cyl. (Door 28)	
28	121	Ambient Temp, to Left-Hand Aileron Act.	
29	122	Case Temp. to Left-Hand Aileron Act.	
		BOX NR 4 (Pressure)	
1		Counter-Time (Omit)	
2		Potentiometer-Time	
3	1	Airspeed	
4	2	Altitude	
5	3	Outside Air Temp.	
6	4	Exhaust Gas Temp	
7	5	Main Fuel Flow (Left-Hand Engine)	(LB/HR)
8	6	Main Afterburner Fuel Flow (Left-Hand Engine)	0-24 GPM
9	8	Tachometer Speed	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 4 (Pressure) (Cont'd)	
10	10	Cabin Conditioning Control Valve Position	0-90° 1000
11	11	Static Reference Manifold	± 6 PSID
12	404	Differential Pressure Between Fwd. Fuel Cell and Fwd. Fuel Cell Cavity	± 7.5 PSID
13	4 50	Total Pressure Cooling Air Inlet to Alternator	± 12.5 PSID
14	451	Alternator, Compartment Static Pressure	± 6 PSID
15	440	Utility Pump Inlet Pressure	± 25 PSID
16	441	Utility Pump Outlet Pressure	0-5000 PSIG
17	442	Flight Control Pump Inlet Pressure	± 2 5 P S ID
18	443	Flight Control Pump Outlet Pressure	0-5000 PSIG
19	444	Flight Control Inlet Pressure, Right- Hand Rudder Act.	0-3500 PSIG
20	430	Static Pressure, Air into Util. Symmetrical Hydraulic Reserve	± 12.5 PSID
21	408	Fuel Pressure at Main Eng. A/B Fuel Manifold Line	0-1000 PSIG
22	421	Fwd. Equip. Compartment Pressure, Differential Pressure to Static Air Sensor	± 6 PSID
		BOX NR 5 (Temperature)	
1	130	AC Alternator Cooling Air Inlet Temp	
2	131	AC Alternator Cooling Air Outlet Temp.	

Table 1 (Cont'd)

BOX POSITION	PICKUP NR	ITEM	RANGE
		BOX NR 5 (Temperature) (Cont'd)	
3	133	Ambient Air Temp, Near Alternator	
4	132	AC Alternator Frame Temp.	
5	60	Fuel Temp at A/B Fuel Flowmeter	
6	134	Frame Temp. of Voltage Regulator	
7	61	Surface Temp. of Stabilizer Act. Bearing	
8	84	Temp, of Air to Pressurize Utility System Hydraulic Reservoir	
9	228	Battery Compartment Ambient	
10	229	Battery Terminal Temp.	
11	76A	Aft Cockpit Air Inlet Temp	
12		High Reference	
13		Bottom Seat Remover	
14		Middle Seat Remover	
15		Top Seat Remover	
16		Canopy Remover	
17		Ambient at Canopy Remover (Spare) Top	
18		Ambient at Canopy Remover (Spare) Bottom	
21		I.C. Compartment	
22		AC Alternator Cool Air Inlet Temp.	
23		AC Alternator Cool Air Outlet Temp.	
24		Alternator Ambient Air Temp	
25		AC Alternator Frame Temp	
26		Aft Cockpit Total Air	

APPENDIX B

DESCRIPTION OF FLIGHTS

GENERAL

Eleven flight tests were made to determine the extreme high temperature operating limits which can be expected to occur throughout the aircraft and to obtain preliminary instrument flight procedures for the aircraft. Temperatures and pressures which occur in the hydraulic oil, engine oil, fuel, and air-conditioning systems were monitored by instrumentation recording equipment to determine their operating limits. In addition to determining extremes, several flights were devoted to normal high temperature operation and design missions to establish an envelope of expected normal desert operating conditions for inclusion in Section IX of the Flight Manual. Preliminary instrument procedures for takeoff, climb, holding, jet penetration, low approach, GCA, and missed approach were also developed during the test period. All but one test flight were flown during the hottest portion of the day. A weather station temperature of 105°F coupled with a solar radiation of 310 btu/sq ft/hr was established as the desired test minimums. The 105°F minimum temperature was obtained on most flights, although, due to the calendar time of the test, the solar radiation was below the desired maximum. A ground temperature survey was conducted prior to each flight which consisted of at least a four-hour heat soak.

Table 2

HIGH TEMPERATURE FLIGHT CONDITIONS

FLIGHT TEST NR 1 (ADVERSE WEATHER PROCEDURES)

DATE: 19 Aug 60 FLT. TIME: 70 Minutes

MAX. AMBIENT TEMPERATURE: 109°F T/O AMBIENT TEMPERATURE: 109°F

OBJECTIVE: To accomplish an inflight temperature survey of various subsystems at an altitude of 20,000 feet, an instrument takeoff (ITO), a ground controlled approach (GCA), and a single engine saw-tooth descent.

COMMENT: A simulated instrument takeoff was made using VFR takeoff procedures. At 20,000 feet the right engine was shut down and a descent made to simulate a single engine penetration. A GCA was simulated utilizing a closed airfield.

FLIGHT TEST NR 2 (MISSION 2, SUPERSONIC TRAINING)

DATE: 22 Aug 60 FLT. TIME: 75 Minutes

MAX. AMBIENT TEMPERATURE: 106°F T/O AMBIENT TEMPERATURE: 106°F

OBJECTIVE: To provide normal systems operating data and to evaluate the air-conditioning and pressurization systems. Adverse weather flight procedures and techniques were practiced at the end of the mission.

Table 2 (Cont'd)

FLIGHT TEST NR 2 (MISSION 2, SUPERSONIC TRAINING) (Cont'd)

COMMENT: An instrument takeoff and a military power climb to 40,000 feet were accomplished. The airspeed was stabilized at .88 Mach for 15 minutes. The aircraft was then accelerated to its maximum power and held for 10 minutes. A holding pattern, descent in a holding pattern, penetrations from 20,000 feet and a GCA were made before returning to the El Centro NALF.

FLIGHT TEST NR 3 (ADVERSE WEATHER PROCEDURES)

DATE: 24 Aug 60 (FIRST FLT.)

FLT. TIME: 90 Minutes T/O ROLL: 2600 Ft

MAX. AMBIENT TEMPERATURE 104°F T/O AMBIENT TEMPERATURE 104°F 3350 Ft LANDING ROLL:

OBJECTIVE: To accomplish a high-speed temperature survey at low altitude, instrument takeoff, two holding penetrations, and two GCA approaches.

COMMENT: An instrument takeoff was conducted in the same manner as previous flights except that A/B operation was maintained longer than before. A holding pattern was performed at 30,000 feet and a penetration was made at an altitude of 20,000 feet. Missed approach procedures were practiced along with two GCA approaches.

FLIGHT TEST NR 4 (MISSION 4, NIGHT FLIGHT)

DATE: 24 Aug 60 (SECOND FLT.)

FLT, TIME: 90 Minutes T/O ROLL: 2550 Ft

MAX. AMBIENT TEMPERATURE: 91°F T/O AMBIENT TEMPERATURE: 91°F LANDING ROLL: 2950 Ft

OBJECTIVE: To perform a temperature survey after sundown. Its purpose was to determine the effects of solar radiation on airconditioning requirements.

COMMENT: The previously developed instrument takeoff and climb techniques were applied to night operations and found acceptable. After climb to 30,000 feet the aircraft was accelerated to and held at its maximum airspeed for 10 minutes. It was then decelerated to and held at its maximum endurance airspeed for 15 minutes. A holding pattern, penetration, and a GCA were performed as in previous flights.

FLIGHT TEST NR 5 (MISSION 1, LOW ALTITUDE)

DATE: 25 Aug 60 (FIRST FLT.) FLT. TIME: 90 Minutes

T/O ROLL 2600 Ft

MAX. AMBIENT TEMPERATURE: 100°F T/O AMBIENT TEMPERATURE: 97°F LANDING ROLL: 3250 Ft

OBJECTIVE: To provide extreme and normal operating data on the various aircraft systems at both high and low altitudes. Penetrations and VOR approaches using GCA procedures were made at the end of the low altitude portion of the test.

Table 2 (Cont'd)

FLIGHT TEST NR 5 (MISSION 1, LOW ALTITUDE) (Cont'd)

COMMENT: After an instrument takeoff and climb to 35,000 feet, power was reduced to best cruise airspeed and held for 10 minutes followed by 15 minutes at maximum endurance airspeed. Following a maximum rate of descent to 700 feet, the aircraft was held at military power at this altitude for 10 minutes. Adverse weather procedures were practiced at the end of the test run.

FLIGHT TEST NR 6 (ADVERSE WEATHER PROCEDURES)

DATE: 25 Aug 60 (SECOND FLT.)
FLT. TIME: 90 Minutes
T/O ROLL: 2600 Ft

MAX. AMBIENT TEMPERATURE: 95°F
T/O AMBIENT TEMPERATURE: 95°F
LANDING ROLL: 3600 Ft

OBJECTIVE: To accomplish an instrument takeoff followed by a single engine penetration, GCA, VOR, and missed approach.

COMMENT: All of the single engine procedures were worked out on this flight. The Yuma VOR and LF stations were used to develop adverse weather procedures.

FLIGHT TEST NR 7 (MISSION 3, HIGH SPEED)

DATE: 29 Aug 60 FLT. TIME: 80 Minutes T/O ROLL: 2600 Ft MAX, AMBIENT TEMPERATURE: 108°F T/O AMBIENT TEMPERATURE: 108°F LANDING ROLL: 3860 Ft

OBJECTIVE: To make inflight temperature measurements of various subsystems at maximum airspeed at an altitude of 20,000 feet. Adverse weather information on flight procedures and techniques were practiced at the end of the test run.

COMMENT: An instrument takeoff and climb to 20,000 feet were made using VFR takeoff procedures. The aircraft was accelerated to its maximum airspeed and maintained for 10 minutes, then decelerated to its maximum endurance airspeed and held for 15 minutes. Two holding patterns and penetrations were performed over the Yuma VOR station. A GCA was flown using VFR procedures.

FLIGHT TEST NR 8 (MISSION 5, LOW LEVEL, HIGH SPEED)

DATE: 30 Aug 60 FLT. TIME: 65 Minutes T/O ROLL: 2700 Ft MAX. AMBIENT TEMPERATURE: 111°F T/O AMBIENT TEMPERATURE: 107°F LANDING ROLL: 3825 Ft

OBJECTIVE: To provide extreme operating data on the various aircraft systems and evaluate the air-conditioning systems at high speed and high ambient temperature conditions.

Table 2 (Cont'd)

FLIGHT TEST NR 8 (MISSION 5, LOW LEVEL, HIGH SPEED) (Cont'd)

COMMENT: An instrument takeoff and climb to 2500 feet were made and the aircraft was stabilized at maximum military power for 15 minutes followed by a climb to 20,000 feet at military power and stabilized at maximum endurance airspeed for 15 minutes. The final portion of the test consisted of a stabilization for 15 minutes at cruise airspeed.

FLIGHT TEST NR 9 (MISSION 6, HIGH ALTITUDE, CRUISE AIRSPEED)

DATE: 2 Sep 60 MAX. AMBIENT FLT. TIME: 105 Minutes T/O AMBIENT T

T/O ROLL: LANDING R

MAX. AMBIENT TEMPERATURE: 106°F T/O AMBIENT TEMPERATURE: 105°F LANDING ROLL: 3530 Ft

OBJECTIVE: To provide simulated design navigational training and normal operational data on all systems of the aircraft. Data was obtained for adverse weather during a portion of this mission.

COMMENT: After an instrument takeoff and military climb to 42,000 feet, the aircraft was held for 30 minutes at its best cruise airspeed (.885M). A descent to 30,000 feet was made and the aircraft held at its best cruise airspeed for 15 minutes followed by a descent to 20,000 feet and a practice penetration for GCA tests and single engine operation procedures.

FLIGHT TEST NR 10 (MISSION 5, REPEAT LOW-LEVEL HIGH SPEED)

DATE: 3 Sep 60 MAX. AMBIENT TEMPERATURE: 115°F FLT. TIME: 60 Minutes T/O AMBIENT TEMPERATURE: 109°F T/O ROLL: 2600 Ft LANDING ROLL: 3950 Ft

OBJECTIVE: To provide data during extreme operating conditions on the various aircraft systems.

COMMENT: An instrument takeoff and military climb to 2500 feet were made. The aircraft was stabilized at maximum military power for 10 minutes, followed by a military-power climb to 20,000 feet, and best cruise airspeed accomplished and held for 10 minutes. Then the aircraft was decelerated to its maximum endurance airspeed and held for 15 minutes.

(See Appendix D For Tabulated Data on this Mission)

WADD TN 60-283

FLIGHT TEST NR 11 (MISSION 7, CRUISE AIRSPEED)

DATE: 4 Sep 60 FLT. TIME: 75 Minutes T/O ROLL: 2600 Ft MAX. AMBIENT TEMPERATURE: 113°F
T/O AMBIENT TEMPERATURE: 110°F
LANDING ROLL: 3400 Ft

(See Appendix D For Tabulated Data on this Mission)

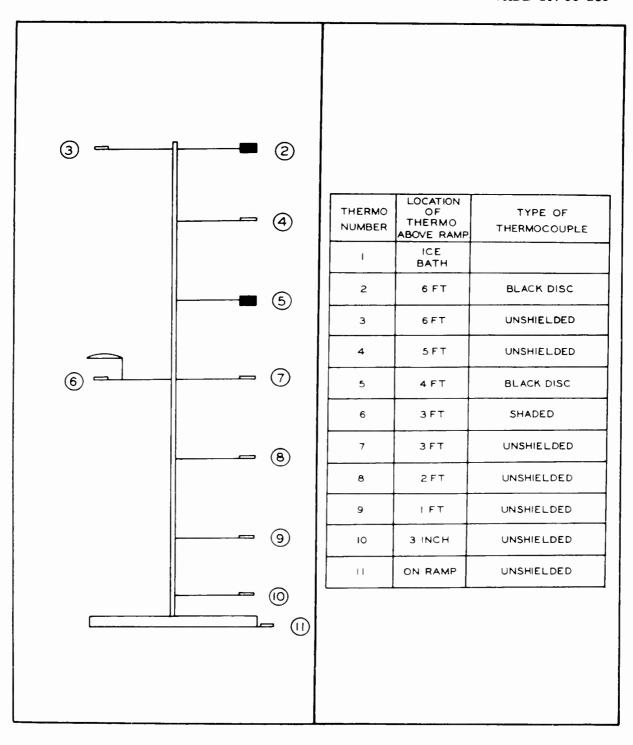
OBJECTIVE: To provide general operating data on the aircraft systems by flying at cruise airspeed at 35,000 and 20.000 feet altitude.

Following an instrument takeoff and military power climb to 35,000 feet, the aircraft was flown for 35 minutes at its best cruise airspeed. A descent to 20,000 feet was made and the aircraft was stabilized at its best cruise airspeed for 10 minutes. Adverse weather flight procedures and techniques were practiced during this mission.

APPENDIX C

STATIC TEMPERATURE SURVEY DATA

Over the weekend of 26-28 August, a static temperature survey was conducted by NALF at El Centro, California, for 60 consecutive hours. The tabulated data is presented in this appendix with selected compartment temperatures and ambient air temperatures presented in graphic form. (Reference paragraphs 17 and 18.)



 $\label{thm:couple_problem} \mbox{Figure 2. Thermocouple Locations for the Static Temperature Survey.}$

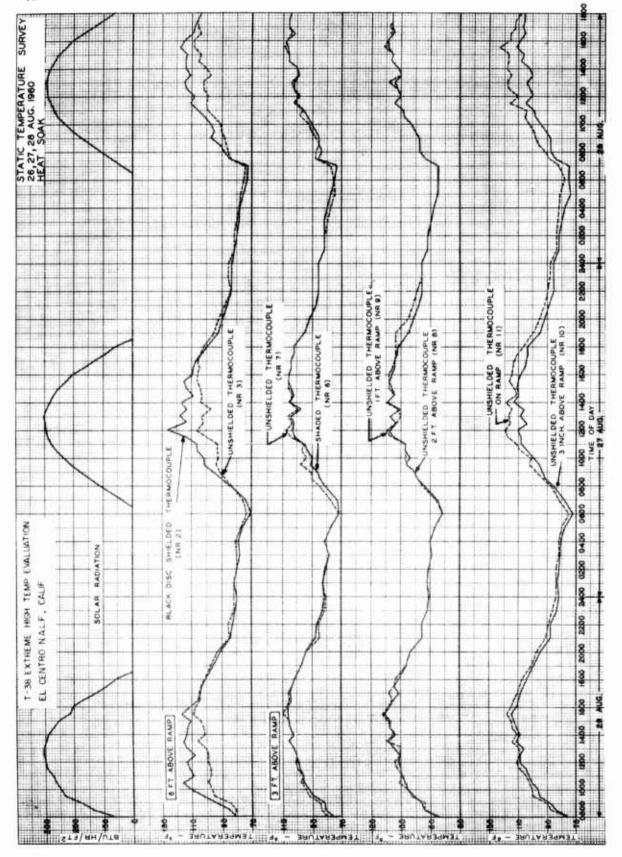


Figure 3. Solar Radiation and Ambient Air Temperatures (26, 27, 28 August 1960).

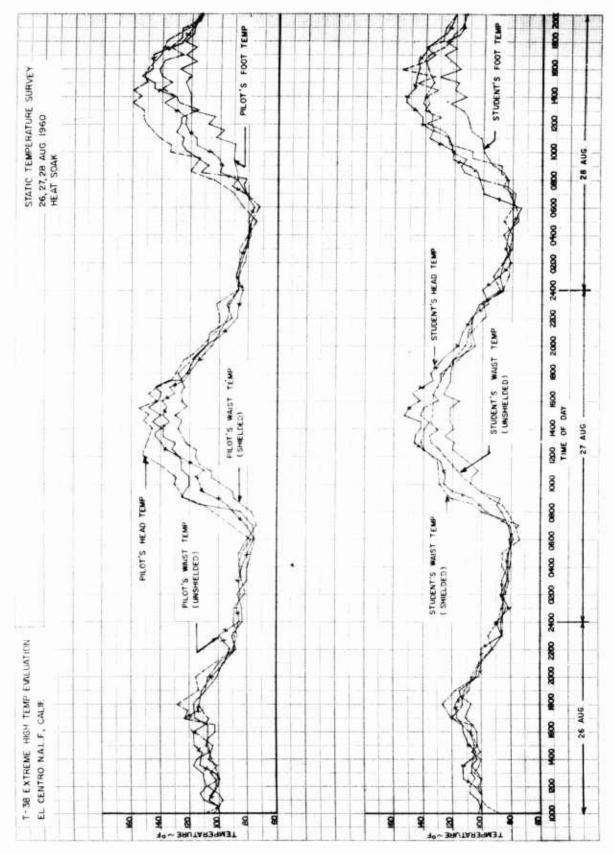
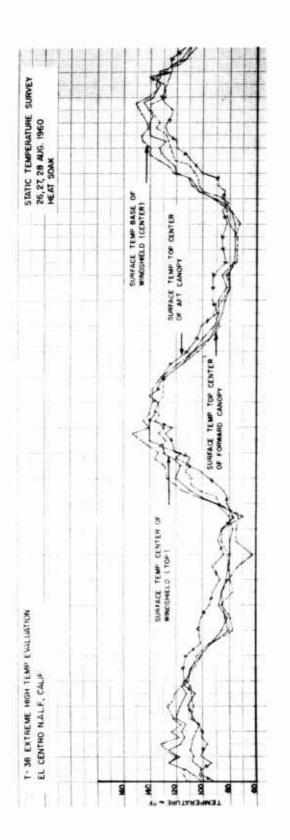


Figure 4. Temperature Survey (Pilot and Co-Pilot).



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Figure 5a. Canopy Surface Temperatures (Interior).

Figure 5b. Escape Systems Actuator Temperatures.

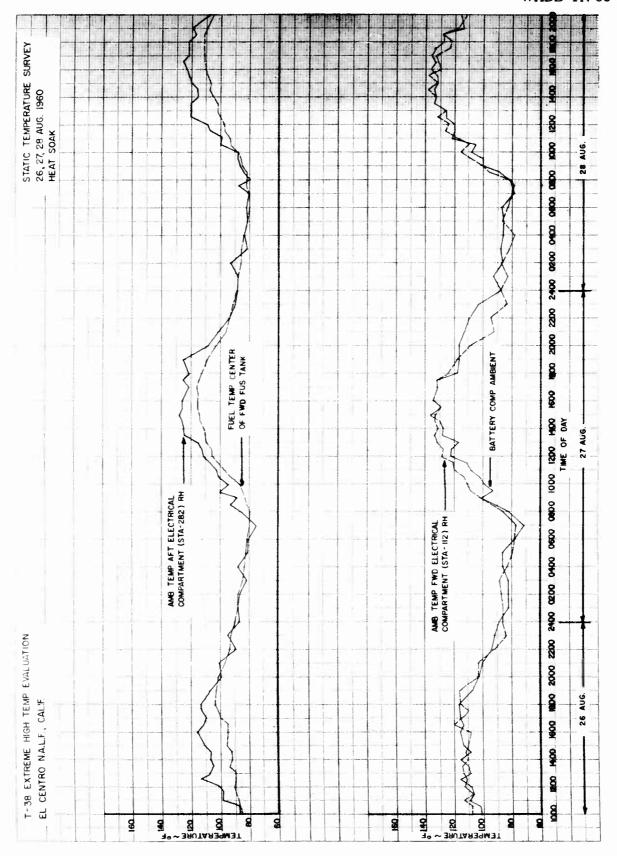


Figure 6. Aircraft Internal Temperatures.

T-38 High Temperature Evaluation Data (26, 27, 28 August 1960)

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2-25 SURF TEMP TEMP CENT CENT CENT 2-24 SURF TEMP TOP CENT FED CAN 2-23 SURF TEMP LEFT SIDE WINDSH T-38 High Temperature Evaluation Data (26, 27, 28 August 1960) 2-22 SURF TEMP CENT TINDSH 2-20 SURF TEMP LEFT CFNT 2-19 SURF TEMP WINDSH RASF LFFT 2-18 SURF TEMP WINDSH PASE CENT 2-17 C.P. C.P. A.I.R. TEMP AIR AIR AIR DNSTR HOT COLD 2-15 AIR AIR ANTI-ICF

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2-26 SURF TEMP LEFT BOTT FWD CAN

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256	292	268	274	280	286	290	296	302	308	314	320	326	332	340	346	352	158	364	370	376	382	388	394	004	907	412	418	424	430	436	244	844	757	460	466	472	478	484	064	967	502	

5-5 FUEL TEMP A/B FLOW METER 5-4 AC ALT FRAME 5-3 AMB AIR AIR ALT 5-2 AC ALT COOL AIR OUT 7 PAPL TEAP PAPL 3-29 CASE TEMP LEFT AILER ACT 3-28 AMB TEMP LEFT AILFR ACT 3-27 SKIN TEMP LFFT AILFR DOOR 3-26 FLT CONT COOL IN OIL TEMP 3-23 FLT CONT RES IN OIL OIL Z

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	126		*	122	118	113	103	93	87	98	4	82		7.30	7.5	7.6	7.3	7.4	11	90	40	87	5	44	101	101	111	113	113	120	122	124	124	125	129	125	124	123	120	118	115	113	
	124		171	120	110	בו	103	93	87	8.7	40	8.2	81	4	7.5	4.	7.3	76	19	82	9.0	90	•	100	103	107	111	11	117	120	120	122	122	124	122	124	123	120	118	117	115	112	
	123		174	124	121	116	105	9.9	90	88		40	82	C	78	7.7	76	16	7.8	7.8	08	83	80	06	46	66	101	106	100	113	114	117	117	120	121	122	124	123	123	122	120	118	
•	126		67	124	122	117	101	9.6	90	9.0	¥ @	83	82	080	7.7	76	73	76	77	7.8	91	86	06	6	96	101	104	100	113	115	118	121	122	122	124	125	126	129	124	123	120	118	
	126	27.	971	123	122	116	101	99	90	88	9.	82	82	0	76	7.7	*	7.5	78	7.8	-	86	90	16	66	100	103	106	113	116	117	120	121	123	124	126	126	125	124	123	120	118	
•	124			123	122	116	101	9.9	91	6	2	•	82	C	4	80	76	11	78	78	00	4	8	C	97	97	103	105	109	112	113	116	118	- A	121	174	124	123	124	122	119	118	
•	1600		00/1	1730	1800	1900	2000	2100	2200	2300	2400	100	200	400	004	200	009	700	730	008	830	006	016	1000	1010	1100	1130	1200	1230	1300	1330	1400	1430	1500	1530	1600	1610	1700	1730	1800	1830	1900	
	256	707	897	214	280	286	290	962	302	308	314	320	326	332	340	346	382	358	364	370	376	382	388	394	400	904	412	410	424	430	436	244	448	484	460	466	472	478	184	490	964	205	

S-16 CAN RFM TEMP 5-15 TOP SFAT REM TEMP 5-14 MID SEAT REM TEMP 27, 28 August 1960) 5-13 GFAT REM TEMP AFT C/P IN TEMP T-38 High Temperature Evaluation Data (26, 5-10 RATT TERM TEMP PATT COMP AMP SURF TEMP STAR ACT ACT 5-6 FRAME TEMP VOLT REG

256	1600	122	124	120	133	129	124	130	86.	140	120
292	1610	122	124	120	132	128	126	130	134	138	128
268	1700	125	122	124	661	127	126	128	131	138	126
274	1710	121	122	122	α [[129	122	126	132	134	125
280	1800	120	120	122	117	126	122	125	127	125	123
286	1000	115	115	115	117	7117	116	119	117	121	117
290	2000	108	101	105	113	107	113	111	114	113	107
296	2100	100	66	101	100	101	106	102	107	104	100
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346	200	C	78	81	7.5	7.8	82	79	9.4	86	78
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388	066	80	0	7.6	100	88	66	26	101	112	63
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004	1030	16	86	103	105	66	101	101	113	118	103
904	1100	101	101	105	122	102	102	107	110	129	109
412	1130	105	106	109	120	108	113	111	120	129	112
418	1200	107	109	110	120	1111	114	115	120	135	115
424	1230	111	115	114	126	116	118	911	127	135	110
063	006	114	114	110	127	119	318	121	132	140	123
436	1330	116	114	122	134	120	120	124	133	141	124
277	1400	117	118	1 B	133	124	118	125	129	140	126
8 7 7	1430	118	1 1 9	122	134	124	125	126	136	145	127
424	1500	119	118	120	132	126	172	128	136	145	128
094	1530	120	120	127	132	128	126	129	132	140	130
466	1600	121	121	122	136	128	126	130	140	130	130
472	1630	122	122	120	134	130	134	130	140	142	129
478	1700	121	121	122	136	129	125	128	135	146	128
484	1730	120	121	122	133	128	133	128	130	141	126
067	1800	120	120	120	128	126	134	126	132	134	124
967	1830	118	118	118	128	124	120	124	123	125	121
205	1900	115	115	115	114	121	118	122	120	133	118
508	2000	110	110	112	115	112	109	113	112	114	111

APPENDIX D

FLIGHT TEST DATA

Selected data from two flights, Flight Test numbers 10 and 11, made by NALF at El Centro, California, September 3rd and 4th, respectively, are presented in this appendix. Flight Test number 10 was conducted at maximum military power at 2500 feet, and at best cruise airspeed and maximum endurance airspeed at 20,000 feet. Flight Test number 11 was conducted at best cruise airspeed at altitudes of 20,000 and 35,000 feet. The static temperature survey (ambient air temperatures and solar radiation for both flights) is presented in graphic form.

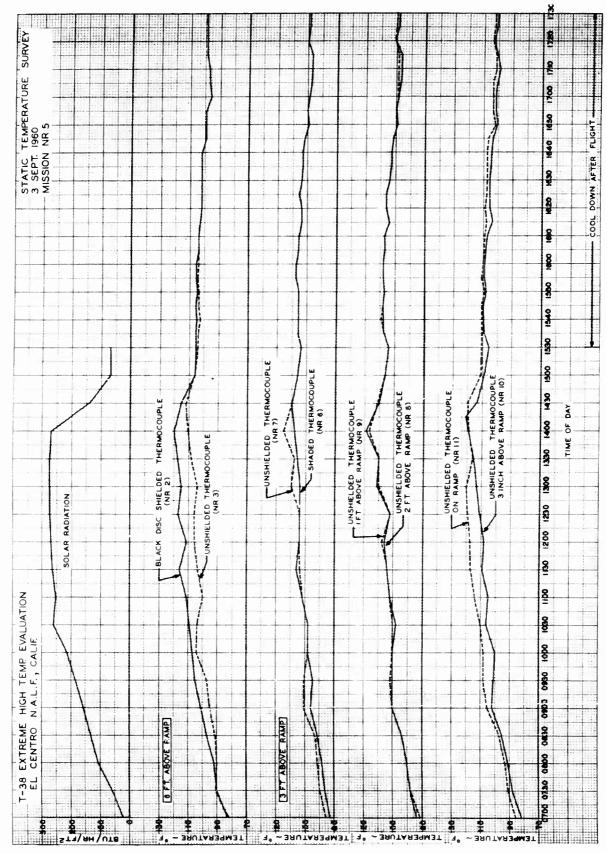


Figure 7. Solar Radiation and Ambient Air Temperatures (3 September 1960).

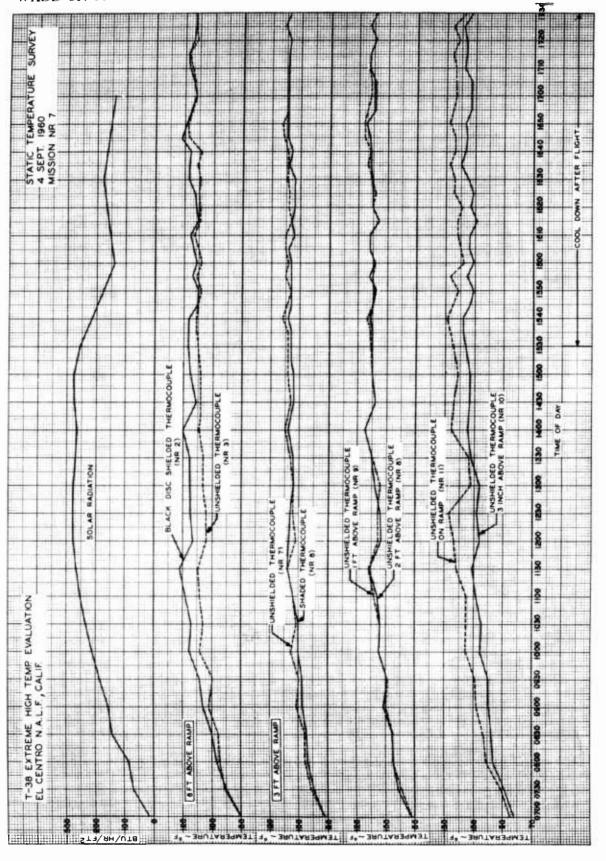


Figure 8. Solar Radiation and Ambient Air Temperatures (4 September 1960).

Time Speed National Property National												
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TIME SPEED TERM TERM PLAN REF AIR PRESS PURP END FINE SPEED TERM FOLLOWS FEEL TERM FLOW PRESS IN THE PRESS PURP END FINE SPEED TERMS PRESS PURP END FINE SPEED	Z	FLAP	AIR	ALT	X	TACH	MAIN	STAT	OWD	FUEL	100	016
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42 570 2080 649 2800 663 6363 2083 446 44 570 2240 620 16286 2800 651 6363 2023 446 46 570 1880 659 659 2017 210 461 460 651 461 465 461 461 460 653 671 46	76	30	570	2110	620	16320	2800	651	6437	2001	011	99
44 570 2240 620 16286 2800 651 6363 2023 446 46 570 1980 620 16347 2800 669 6350 2071 461 48 565 2110 620 16286 2800 649 6375 2071 461 67 560 2310 620 16286 2800 649 637 2119 461 64 560 240 620 16218 2800 649 637 2102 462 68 560 2410 620 16218 2800 643 637 2102 462 70 560 2410 620 16150 2800 649 637 2102 462 70 560 2410 620 16150 2800 649 650 2124 471 70 560 2410 620 16280 2800 649 637 <td>78</td> <td>4.2</td> <td>570</td> <td>2080</td> <td>620</td> <td>16490</td> <td>2800</td> <td>663</td> <td>6350</td> <td>2089</td> <td>450</td> <td>06</td>	78	4.2	570	2080	620	16490	2800	663	6350	2089	450	06
46 570 1980 620 16437 2800 609 6350 2045 461 60 566 2310 620 16286 2800 643 6375 2045 450 67 560 2310 620 16286 2800 663 6373 2141 460 64 560 2440 620 16218 2800 663 637 2141 460 68 560 2410 620 16150 2800 651 6437 2102 462 70 2410 620 16150 2800 651 6437 2102 462 70 2410 620 16150 2800 651 6437 2102 462 70 250 2620 16150 2800 651 6437 2102 462 70 250 2620 16150 2800 650 651 650 2800 650 6	80	77	570	2240	620	16286	2800	651	6363	2023	446	69
4.8 565 2110 620 16218 2800 639 6375 2045 450 6.0 560 2310 620 16490 2800 649 637 2159 453 6.4 560 2440 620 16218 2800 643 5124 461 6.6 560 2440 620 16456 2800 633 6227 2124 461 70 560 2570 620 16456 2800 639 6350 2102 462 70 560 2510 620 16456 2800 639 6350 2102 462 70 2570 620 16526 2800 663 6412 664 471 70 2570 620 1656 2800 663 6412 671 468 80 565 2410 620 16354 2800 663 6412 471 80<	82	46	570	1980	620	16337	2800	609	6350	2071	461	120
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Z	FLAP	016	DIFF	STAT	TOTAL	STAT	TOTAL	TOTAL	STAT	STAT	0
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		T()O	CVP	RLEFD	BLEED	CNOC	COND	RLEED	DEF	CIP	FUEL
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			SFNS	3-05	S-0FF	HOT	HOT	PASS	20N	VAL	4
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89	22	1657	2128	567-	1153	382-	62	935	62	669	223
10	24	1818	2141	525-	1124	355-	5.7	076	62	669	2250
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74	28	1550	2200	579-	1170	389-	56	776	4	111	2273
76	30	1899	2148	710-	1174	375-	47	676	79	711	2277
78	4.2	1720	2154	525-	1133	396-	54	076	77	722	2325
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138	164	2096	3193	238-	485	104	352	383	241	286	2857
140	174	2087	3154	131-	867	-0	338	394	248	229	2700
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4-22 FWD COMP COMP $\begin{array}{c} \mathbf{x} \bullet \\ \mathbf{x} \bullet \mathbf{x} \bullet$ 4-21 FUEL PRESS MAIN A/3 LINE PRESS PRESS PRESS AIR CTIL SYX PRO PRICO T-38 High Temperature Evaluation Data, Flight Nr 10 (3 September 1960) FLT CONT IN PRESS R-RUD ACT FLT CONT PUMP OUT PRESS FLT CONT PUMP IN PRESS 4-16 UTIL PUMP OUT PRESS 4-15 UTIL PUNP IN PRESA 4-14 ALT COMP STAT PRESS 4-13 TOTAL PRESS COOL AIR IN ALT ELAP TIME 3

WADD TN 60-283

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2-12 DEFOG AIR TEMP UPSTR 2-11 C/P AIR OUT DNSTR FOUIP RAM 2-10 AMB TEMP ADJ CABIN SENS T-38 High Temperature Evaluation Data, Flight Nr 10 (3 September 1960) 2-9 01L 01L 0UT FNG 2-8 01L 1FKP 1N ENG 2-7 FUFL TEMP IN OIL 2-6 FUEL TFMP O1L COOL 7-4 FUFL TEMP OUT FWD CFLL ROOST 2-3 FUFL TEMP CENT FWD FUS TANK 2-2 COMP AIR IN TEMP 7.000 2-1 COMP AIR IN TEMP 11.00 ELAP TIME Z

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2-23 SURF TEMP LEFT SIDE WINDSH 2-22. SURF TEMP CENT WINDSH 2-21 SURF TEMP LEFT CENT EFFT SIDE T-38 High Temperature Evaluation Data, Flight Nr 10 (3 September 1960) 2-20 SURF TERP CERT CERT TINDSH 2-19 SURF TEMP WINDSH RASE LFFT SIDF 2-16 AIR AIR DNSTR HOT COLD 2-15 A M P A D D N S T P A N S T P A N T I P A 2-14 BLEED AIR AY PASS 2-13 RLEED AIR TEMP DNSTR S-OFF $\begin{array}{c} \mathbf{v}_{1} \otimes \mathbf{v}_{2} \otimes \mathbf{v}_{3} \otimes \mathbf{v}_{4} \otimes \mathbf{v}_{3} \otimes \mathbf{v}_{4} \otimes \mathbf{v}_{3} \otimes \mathbf{v}_{3} \otimes \mathbf{v}_{4} \otimes \mathbf{v}$

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T-38 High Temperature Evaluation Data, Flight Nr 10 (3 September 1960)

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5-13 BOTT SEAT REM TEMP S-11 AFT C/P IN S-10 RATT TERM TEMP I-38 High Temperature Evaluation Data, Plight Nr 10 (3 September 1960) 5-9 (0MP AMB TEMP ATEMP PRESS UTIL SYN HYD RES SURF SURF TEMP STAR ACT REAR 5-6 FRAME TEMP VOLT REG FUEL TEMP A/B FLOW METFR 3-29 CASE TEMP LEFT AILER 3-28 AMB TEMP LEFT AILER FLAP Z

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	T	T-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960)	emperatu	re Evalu	ution Data	, Flight N	r 11 (4 Se	ptember	(0961		
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36		350	777	15640	006	5664	2773	2255	213	1551
i.		500	777	15470	006	2664	2513	2255	217	1661
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1-38 HIGH TEMP EVALUATION FL CENTRO NALE. CALIFORNIA 4 SEPTEMBER 1960 T-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960) FLIGHT TEST NR 11

		1-18	1-70	1-21	1-22	1-24	1-24	1-76	1-21	1-24	4-12
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152	410	2607	5831	113-	298	10	507	239	155	140	1665
154	420	2688	5844	161-	278	20-	501	545	260	321	1500
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74	520	2759	5805	149-	290	-06	503	235	552	183	832
76	530	2598	5850	-69	261	34-	503	244	437	404	000
7.8	240	2446	5877	137-	546	7	516	235	564	553	727
80	950	3145	5857	119-	356	27-	500	282	556	195	795
82	260	2956	5824	197-	340	-69	522	282	999	506	7.80
88	290	2222	3204	191-	402	104-	247	408	275	307	009
06	009	2329	3349	244-	498	7	429	369	275	458	505
85	610	2329	3271	119-	694	76	368	383	275	424	525
76	620	2114	3258	328-	867	13	374	408	266	263	577
96	630	2329	3232	232-	867	20	395	413	285	344	472
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4-21 4-22 FUEL FWD PRESS EQUIP MAIN COMP A/B D.P.	516 PSID • XXX 1 234 5 149	2 1 0 1 0 1 0 1 0 1 0 1 0 1 0 1 0 1 0 1		1		3 3 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5
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4-14 ALT COMP STAT PRESS	PS10 *XXX 149-	21 21 71 71 78 71	270 270 142- 199- 213-	263- 263- 213- 244- 213-	1117 1117 1117 1117 1117 1117 1117 111	235- 235- 235- 235- 255- 255- 255- 255-
4-13 TOTAL PRESS COOL AIR IN ALT	A 40 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	000 000 000 000 000 000 000 000 000 00	70000 00000 00000 00000	744 404 744 744 744 744 744 744	6 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4
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T-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960)

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356	454	747	348	760	348	34.2	345	360	366	369	185	386	r (0	10 K	350	362	416	398	437	4 3 3	400	0.47	064	767	437	624	424	4 7 00 0 0 0	6/4	531	561	536	877	977	4.50 6.5.3	104	367	352	411	4 C C	000	311	283	417	258	234	238	206	194	170	167	č
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2-23 SURF TEMP LEFT SIDE WINDSH AFT	115 115 121 138 138 127 128 126 126	
2-22 SURF SURF TEMP CENT TOP	04 98 98 98 98 98 98 98 98 98 98 98 98 98	
2-21 SURF TEMP LEFT CENT WINDSH LEFT SIDE	10000000000000000000000000000000000000	
2-20 SURF TEMP LEFT CENT WINDSH		
2-19 SURF XINDSH BASE LEFT SIDE	1111 111111111111111111111111111111111	
2-18 SURF FIRMP WINDSH CENT	0000 04000004004	
FWD C/P C/P AIR IN	000000000000000000000000000000000000000	
2-16 AIR TEMP DNSTR HOT COLD	00110011110011111001111100111110011111001111	
2-15 AIR TEMP DNSTR MOIST SEP ANTI- ICE		
2~14 BLEFD AIR BY PASS TEMP	122 122 122 122 123 123 123 123 123 123	
2-13 BLEED AIR TEMP DNSTR S-OFF	1000 1000 1000 1000 1000 1000 1000 100	
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		T-38	High Tem	T-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960)	Evaluatio	n Data, F	light Nr 1	1 (4 Septe	mber 196	(0		
₹	FLAP	2-24 SURF	2-25 SURF	2-26 5URF TEMP	2-27 SURF TEND	2-28 5URF	2-29 SURF	3-1 P1L0T	3-7 P1L0T	3-3 PILOT	3-4 P1L0T	3-5 STUD
		TOP	LEFT	LEFT	100	LEFT	F F F	TEMP	- CL	TEMP	HEAD	TEMP
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00	0.7	117	71.		701	/ 01	101	513	100	111	107	109
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99	90	111	118	114	101	103	100	112	112	113	100	112
68	09	109	116	118	9.5	102	107	115	109	6	110	117
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134	320	47	28	35	115	118	118	69	6 0	52	107	69
136	330	بر ب	32	43	107	122	113	63	7.1	59	100	63
138	340	40	32	30	107	109	=	76	6.8	97	101	73
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T-38 High Temperature Evaluation Data. Flight Nr 11 (4 Sentember 1960)

3	ELAP	3-6 STUD	3-7 STUD	3-8 STUD	3-10 AMB	3-11 AMB	3-12 AMB	3-13 AMB	3-14 AMB	3-15 AMB	3-16 AMB	3-17 UTIL
	TIME	WAIST	WAIST	HEAD	TEMP	TEMP	TEMP	TEMP	TEMP	TEMP	TEMP	PUMP
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					282	282	277	103.5	112	135	135	
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26		105	107	115	115	107	108	126	146	117	125	101
58	10	101	107	115	113	113	108	129	140	115	126	107
20	20	110	101	109	112	110	102	126	136	120	124	109
25	30	108	108	102	115	113	109	133	138	115	126	114
*	0 0	109	103	107	110	114	113	127	139	117	127	115
0 40	0 4	601	8 6	501	0 0	104	109	128	136	117	129	120
0 0	20 40	0	201	101	101		104	120	9 7	171	136	122
	3 4	103	1 00	0.0	116	711	000	0 4	11.	911	127	9 .
4	40	6	69 0	607	107	117	114	2.0	138	112	126	122
9,	90 90	96	6	76	120	113	-	11.6	111	100	120	071
. 8	10	92	86	96	123	114	126	127	- 04		121	136
9	0.8	06	91	6	118	122	119	117	128	101	128	120
32	06	46	88	66	117	118	115	131	128	101	125	130
1	100	63	90	8 0 6 0	117	121	115	115	128	102	114	128
90	110	80 F	0 0	06	109	107	107	66	116	105	113	125
0 0	120	, o) C	£ 0	4 6	105	101	103	112	101	112	120
2 2	140	101	8	86	7	4	000	9 4	701	£ 0	6 6	112
4.	150	88	80	06	00	98	68	62	101	080		
96	160	76	7.8	98	61	7.4	65	67	80	57	81	86
6 0	170	92	78	ec 60	\$0	69	7.5	5.5	85	26	73	96
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. 4	761	60 60	. 4 4	C 6	9 4	4 4	90	4 6	06	69	69	80
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2	198	88	73	6 0	9	73	98	20	76	2	8.6	2000
2	210	47	74	91	57	67	80	34	6	61	99	8
4	220	90 ·	6,	986	64	63	73	1	80	54	59	80
0 =	240	# C	è 5	* C	25	919	1,	1,	9,	09	6 9	73
0	250	7.0	99	7 60	C 4	, r) v	4	200	6.4	9:	. 3
2	260	16	99	6	4	, v	Ç 4	4	7 1	C 7	. 4	6
4	270	76	69	73	F 7	ec er	99	3.5	9	4 4	9 6	- 4
9.	280	67	9	82	43	67	62	97	61	42	, v.	9.6
6 0 5	290	89	57	CB	4 5	75	o u	18	63	67	53	59
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25	320	- ac	0 ec	9 6	96	- 6	4 (0.5	20	38	56	61
	330	9 4	, r	7.		70	70	7.	25	2 6	64	6]
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9	350	89	9	7.5	37	55	8 7	30	64	. 4	4 4	2.0
2	340	4.7	9	;	,							2
•	200	0	13	-	26	7	42	7	6 0 7	07	01	ď

3-17 UTIL PUMP IN OIL TEMP	141	138	138	136	135	134	132
3-16 A 448 TEMP FWD COMP RH 135	128	133	140	135	136	153	134
3-15 AMB TEMP FWD FLECT COMP LH	145	135	207	201	151	147	141
3-14 AMB TEMP FWD FLFCT COMP RH	155	178	161	174	171	167	170
3-13 AMB TEMP FWD FLFCT COMP RM	181	198	193	196	186 196	187	188
3-12 AMR TEMD AFT COMP STA	113	112	811	111	111	116	120
3-11 AMB TEMP AFT COMP LH	177	111	144 120	117	126	118	120
3-10 AMR TEMP AFT COMP RH	112	119	115	117 121	119	120 125	119
3-8 HEAD TEAD	106	117	118	116	117	114	113
3-7 STUD MAIST TEMP SH	108	112	113	124	115	115	120
3-6 STUD WAIST TEMP UNSH	109	110	115	111	113	111	114
ELAP TIME	1270	1340	1520	1620 1670	1720	1800 1850	1910 1970
z U	268	272	276 278	280 282	286	288	292

		1-56 righ temperature rvatuation bada, Fight Nr 11 (4 September 1960)	-								
		3-18	3-19	3-20	1-21	3-22	3-23	3-24	3-25	3-26	3-27
ž	ELAP	UTIL	UTIL	U T 1 L	FLT	FLT	FLT	FLT	1	-	SKIN
	TIME	PUMP	RES	1000	CONT	LNOU	CONT	LNOU	CONT	LOO	FEE
		001	z E	Z	d∧∩d	d M∩d	RES	Z.	100	C00L	1 6 7
		011	016	011	Z	100	Z	0.71	011	2	ATLER
		F	TEMP	TEMP	016	OIL	OTL	TEMP	TEMP	011	DOOR
					3	d. Fu.	TEMP	R-PUD	R-RUD	TEMP	28
								7	7		
	×										
3 6		111	113	107	107	118	112	00	a	116	
58	10	120	115	115	107	131	120	120	123	120	120
09	20	128	122	120	107	140	120	130	127	124	120
62	30	130	125	128	115	142	118	133	134	131	118
49	04	140	122	132	118	147	121	142	140	134	120
99	50	141	128	137	120	157	124	146	148	145	115
89	09	151	130	143	122	163	128	154	155	149	117
0 1	62	150	138	143	122	163	130	154	154	148	116
2.5	4	142	128	14.2	124	162	129	151	154	146	118
	0	# C	171	241	9 2 3	161	128	14.9	153	144	114
9 0	0 6	14	100	# F	122	166		151	157	147	115
0 0		100	13.5	4 7	122	166	133	151	157	150	118
82	0 6	157	136	 	128	0 4	1 2 3	101	000	0 5 1	120
4 6	100	158	134	15.5	131	171		1 1	15.4	101	120
98	110	157	130	148	125	173	129	0 5	157	7 1	0
8.6	120	154	120	148	122	167	118	146	155	150	
06	130	144	115	146	118	167	112	144	153	146	71
26	140	142	113	54.	117	166	112	138	152	145	61
•	150	/ E [102	140		15.5	107	136	147	140	4
	007	123	0 00	E .	101	151	6.0	128	140	138	39
00	180	120	e e	131	0 0	140	2 6	120	7 7 7	134	3.6
02	190	118	86	124	60	9 60	82		150	1 2 4	0 0
*0	192	110	7.8	127	96	136	e0 e0	117	136	27.	22
90	194	117	8 \$	128	66	138	9.5	115	138	128	23
80 F	196	115	82	128	76	136	88	111	131	124	23
01	198	115	7	126	88	131	98	113	128	122	25
12	210	109	92	118	9 9	133	4 8	105	126	124	23
	022		1	110	98	126	0 80	105	127	120	18
0 e	240	001	7	100	3 C 10 ■	171		101	122	117	16
20	250	100	67		7.0	120	0 00) r	115	= :	2:
22	260	96	67	6	73	8 -	36	60		2 5	•
24	270	66	62	86	7.	116	74	60	100	105	11
56	280	66	63	7 0	49	116	73	9 8	104	100	12
9 6	062	80 v	69	100	99	109	79	26	102	86	Ξ
		0 0	0 5		20	107	57	87	105	100	•
3.4	9.50	9 4	9 40	. 6	4 4	111	U (98	101	95	4.
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3 6	046	60	63	•	9 0		2,5	c o	8 6	6	۲.
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24	160	90	57	66	. F	107	v •€	6	* 0	· 6	0 4
44	370	6 0	99	6	9	108	? 4	60	9.	9	- ۱
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75-5	SKIN	TEMP	1 5 5 1	ATIFE	000G	28			121	120	120	27	133	122	122	120	121	4 -	9 .	911		110	911
3-26	1	LNC	000	Z	011	TEMP			126	122	122	121	122	122	121	122	123	122	122	122	126	122	771
3-24	1	LNOU	100	0.11	TEMP	R-RUD	7		182	180	179	7 4	173	171	0 4	168	167	165	161	130		157	
3-24	FIT	LNOU	Z	011	TEMP	R-RUD	CYL		182	181	17.1	17	173	171	170	167	166	163	161	161	140	157	
3-23	FLT	CONT	RES	z	016	TEMP			122	122	126	121	120	122	122	122	124	124	124	124	124	124	
3-22	FLT	TNOU	d₩∩d	001	016	TEMP			129	123	122	122	122	121	122	121	122	122	122	122	122	123	
3-21	FLT	CONT	PUMP	Z	011	TEMP			146	146	144	144	143	144	142	142	144	142	145	141	140	140	
3-20	UTIL	7000	Z	016	TEMP				115	116	135	3 3	114	113	113	123	115	1:2	111	111	112	111	
3-19	UTIL	RES	z	011	TEMP				116	125	117	118	111	120	117	115	113	117	111	119	113	119	
3-18	UTIL	d M∩d	TOO	011	TEMP				118	117	1117	118	116	114	115	113	114	113	113	113	111	111	
	FLAP	TIME						γ•	1270	1300	1340	1400	1470	1520	1620	1670	1726	1770	1800	1850	1910	1970	
	2								268	270	272	274	276	278	280	282	284	286	288	290	262	767	

5-13 BOTT SEAT REM TEMP 5-11 AFT C/P IN TEMP 5-10 BATT TERM TEMP $\begin{array}{c} \text{$0$} \text{$ F-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960) 5-9 COMP AMB TEMP A18 A18 TEMP PRESS UTIL SYN HYD RES SURE SURE STAB ACT BEAR 5-6 FRAME TEMP VOLT REG 5-5 FUEL TEMP A/B FLOW METER $\begin{array}{c} \mathbf{z} = \mathbf{z} \\ \mathbf{z} = \mathbf$ 3-29 CASE TEMP LEFT AILER 3-28 AMB TEMP LEFT AILER TIME S

0.4 & 0.0 &

		3-28	3-29	5-5	2-6	5-7	5-8	6-6	5-10	5-11	
Š	ELAP	AMB	CASE	FUEL	FRAME	SURF	AIR	PATT	BATT	AFT	
	TIME	TEMP	TEMP	TEMP	TEMP	TEND	TEMP	dwo)	TERM	d/)	
		LEFT	LEFT	A / 8	VOLT	STAB	PRESS	AMB	TEMP	Z	
		AILER	AILER	FLOW	REG	ACT	UTIL	TEMP		D. H.	
		ACT	ACT	METER		BEAR	SYN			ı	
							HYD				
							RES				
	*										
266	1210	122	122	164	100	134	120	141	104	0	
268	1270	122	122	169	109	133	Ξ	142	801	103	
270	1300	122	126	161	109	132	120	64	001	000	
272	1340	123	122	156	111	133		142	001	104	
274	1400	125	122	153	109	132	111	154	113	104	
276	1470	124	120	152	110	132	126	132	113	114	
278	1520	124	129	151	109	130	120	166	114	112	
280	1620	122	128	150	110	132	113	159	115	105	
282	1670	125	125	148	111	132	115	166	115	=	
284	1720	124	119	841	111	111	112	154	116	120	
286	1770	122	126	1 # K	111	128	114	150	117	101	
288	1800	124	120	146	111	130	115	155	117	113	
290	1850	122	121	144	112	130	110	157	120	111	
262	1910	122	118	144	111	129	115	14.8			
767	1970	121	119	142	111	128	114	183	110		

S-26 AFT C/P TOTAL AIR TEMP 5-25 AC ALT FRAME AMB AMB AIR AIR ALT T-38 High Temperature Evaluation Data, Flight Nr 11 (4 September 1960) 5-23 AC ALT COOL AIR OUT 5-22 AC ALT COOL AIR IN 5-18 AMB CAN REM TEMP $\begin{array}{c} \text{condoco} \\ \text{condoco}$ AMB AMB CAN REM TEMP 5-16 CAN REM TEMP 5-15 TOP SEAT REM TEMP SEAT REM TEMP ELAP TIME č

 $\begin{array}{c} \mathsf{CIRR} \mathsf{COM} \mathsf{COM}$

ELAP MID TOP CAN AMR AMB AC AC AMB ACT TIME SEAT SFAT PFM CAN ALT ALT TEMP TEMP TEMP TEMP TEMP TEMP TEMP TEM		5-14	5-15	5-16	5-17	5-18	5-22	5-23	5-24	5-2E	5-26
SEAT SFAT PFW CAN ALT ALT ALT ALT ALT ALT TEMP FFM COOL COOL TEMP FFM COOL TEMP ALT TEMP	FLAP	E	TOP	AA	AMR	AMB	AC	V	AMB	V	V 1
REM TEMP REM PEM COOL COOL TEMP ALT	TIME	SEAT	SFAT	3 Li d	AA	₽ ¥ U	ALT	A! T	AIR	ALT	0/
TEMP TEMP TEMP AIR AIR AIR ALT TOP ROTT TEMP AIR AIR ALT TOP ROTT TEMP TEMP TEMP TEMP TEMP TEMP TEMP TE		RFA	RE	TEMP	REM	N Ld	1000	1000	TEMP	FRAME	TOTAL
TOP ROTT IN OUT TEMP TEMP 107 109 111 110 109 123 162 136 109 109 117 118 115 132 154 130 107 117 120 112 111 126 147 139 117 114 125 115 117 136 148 132 117 114 125 126 117 136 148 132 118 115 127 115 116 117 136 148 137 119 117 137 126 114 136 146 137 119 117 137 126 114 136 146 127 111 126 115 110 130 146 127 111 126 115 110 130 130 138 111 126 115 117 131 131 131 110 111 126 115 110 130 140 128 111 12 126 115 117 140 133		TEMP	TEMP		TEMP	TEMP	AIR	A TP	AL T	TEMP	A
TEMP TEMP 107 109 111 110 109 123 162 136 109 112 120 114 116 136 159 141 100 115 120 115 111 126 148 132 117 114 124 120 117 118 118 126 148 118 116 116 124 120 117 136 148 119 116 127 116 117 134 140 119 118 129 121 123 124 146 137 111 126 115 110 130 130 110 111 126 115 110 130 130 110 111 126 115 110 130 130 111 126 115 110 130 130 111 126 115 110 130 130 111 126 115 110 130 130 111 127 128 130					100	BOTT	Z	FIG	t		F
107 109 111 110 109 123 156 107 109 117 118 115 136 136 107 113 120 114 116 136 149 107 113 120 115 111 126 147 117 114 124 136 139 113 115 116 117 136 136 113 115 116 117 134 147 113 116 127 116 117 136 136 113 116 127 126 136 126 113 116 127 126 136 136 113 126 127 146 126 113 126 115 117 126 138 113 126 117 117 126 138 113 126 126 136 136 113 126 127 144 136 113 126 126 138 136 113 126 126 138 136 113 126 126 138 <td></td> <td></td> <td></td> <td></td> <td></td> <td></td> <td>TEMP</td> <td>TEMP</td> <td></td> <td></td> <td></td>							TEMP	TEMP			
107 109 111 110 109 123 162 136 109 109 117 118 115 136 139 162 136 109 107 113 120 114 116 136 147 139 141 117 114 126 147 139 141 117 114 126 147 139 131 118 118 118 118 118 118 118 118 118 118 118 126 146 127 131	×										
109 109 117 118 115 132 154 130 107 113 120 114 116 136 159 141 117 113 125 115 117 136 148 132 117 114 124 127 115 117 134 145 118 115 127 115 117 134 146 137 119 111 126 115 110 130 146 127 111 111 126 115 110 130 140 113 120 126 126 131 131 113 120 126 126 131 131 114 120 126 126 134 134 115 120 126 126 134 134 117 120 126 126 136 136 118 120 126 126 136 136 119 110 120 126 126 136 110 111 120 126 126 136 111 120 120 120 120 112 120 120 120 120 113 120 120 120 120 114 127 127 141 131 115 117 127 141 131 117 117 127 141 131 118 119 120 120 120 119 110 120 120 110 111 120 120 110 111 120 120 110 111 120 120 111 112 120 120 111 112 120 120 111 112 120 120 111 112 120 120 111 120 120 111 120 120 112 120 120 113 120 120 114 120 120 115 120 120 117 120 120 118 120 120 119 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 110 120 120 120 120 120 120 120 120 120 120 120 12	1270	107	109	111	110	109	123	162	136	153	111
107 113 120 114 116 136 159 141 170 115 117 136 147 139 141 171 134 134 135 137 134 134 135 137 134 134 135	1300	109	109	1117	118	115	132	154	130	64	113
109 115 120 112 111 126 147 139 137	1340	107	113	120	114	116	136	159	141	8 7 1	118
117 113 125 115 117 136 148 132 117 114 124 126 126 131 145 145 145 113 115 116 116 137 136 137 136 137 136 137 136 137 136 137 136 137 136 137	1400	109	115	120	112	111	126	147	139	148	117
117 114 124 120 124 131 145 147 147 118 118 126 137 118 118 126 137 131	1470	1117	113	125	115	117	136	148	132	144	127
113 115 127 115 118 126	1520	117	114	124	120	126	131	1.5	142	144	121
116 116 124 116 117 134 146 137 138	1620	113	115	127	115	118	126	•	•	142	14.2
113 117 132 126 114 136 142 128 115 116 124 124 134 134 131 131 131 134 134 137 131	1670	116	116	125	116	117	134	146	137	142	122
115 116 124 126 117 • 146 134 134 133 118 129 121 127 124 146 127 131	1720	113	117	132	126	114	136	142	128	142	114
	1770	115	116	126	126	1117	•	146	134	140	128
111 126 115 110 130 140 128 136 110 110 110 130 140 128 136 110 110 1110 112 126 138 136 131 120 126 126 115 127 143 133	1800	113	118	129	121	123	124	146	127	138	122
110 117 125 111 112 126 138 136 136 137 143 133	1850	113	111	126	115	110	130	140	128	138	129
113 120 126 126 115 127 143 133	1910	110	113	125	113	112	126	138	136	138	124
	1970	113	120	126	126	115	127	143	133	135	126

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